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Mission Objectives

- Find and characterise other Earths orbiting other stars
- Provide unique imaging capability for general astrophysics

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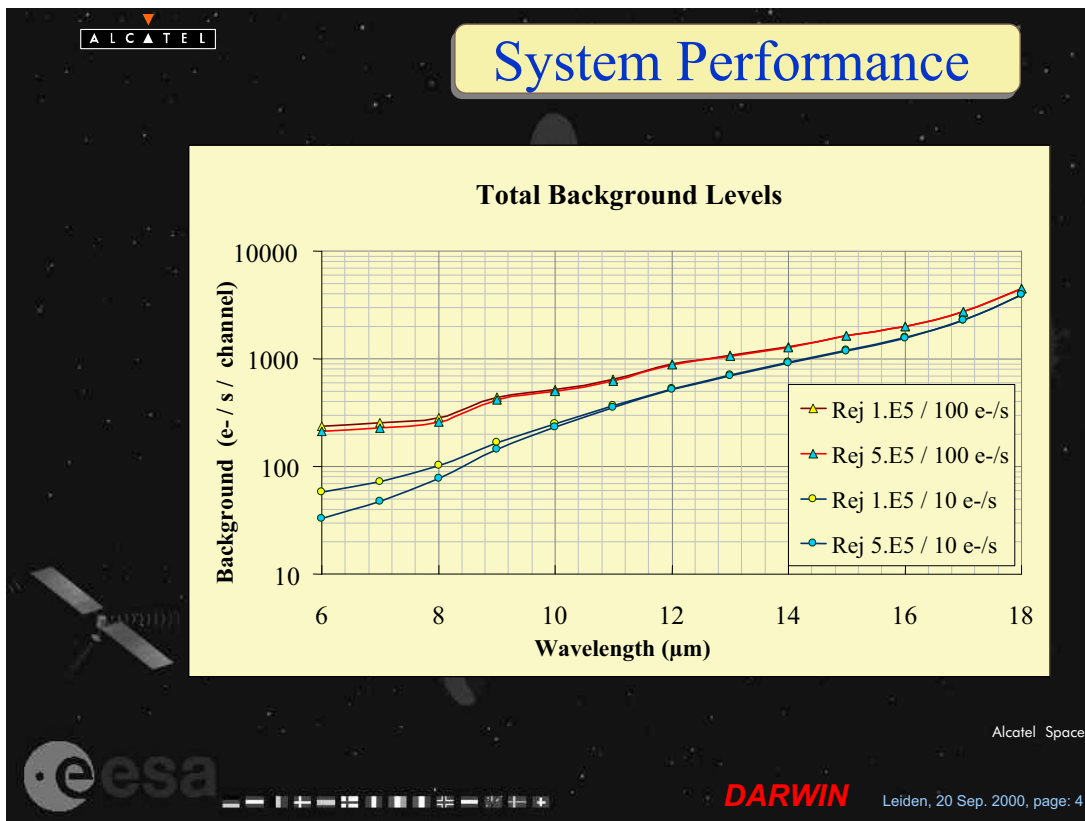
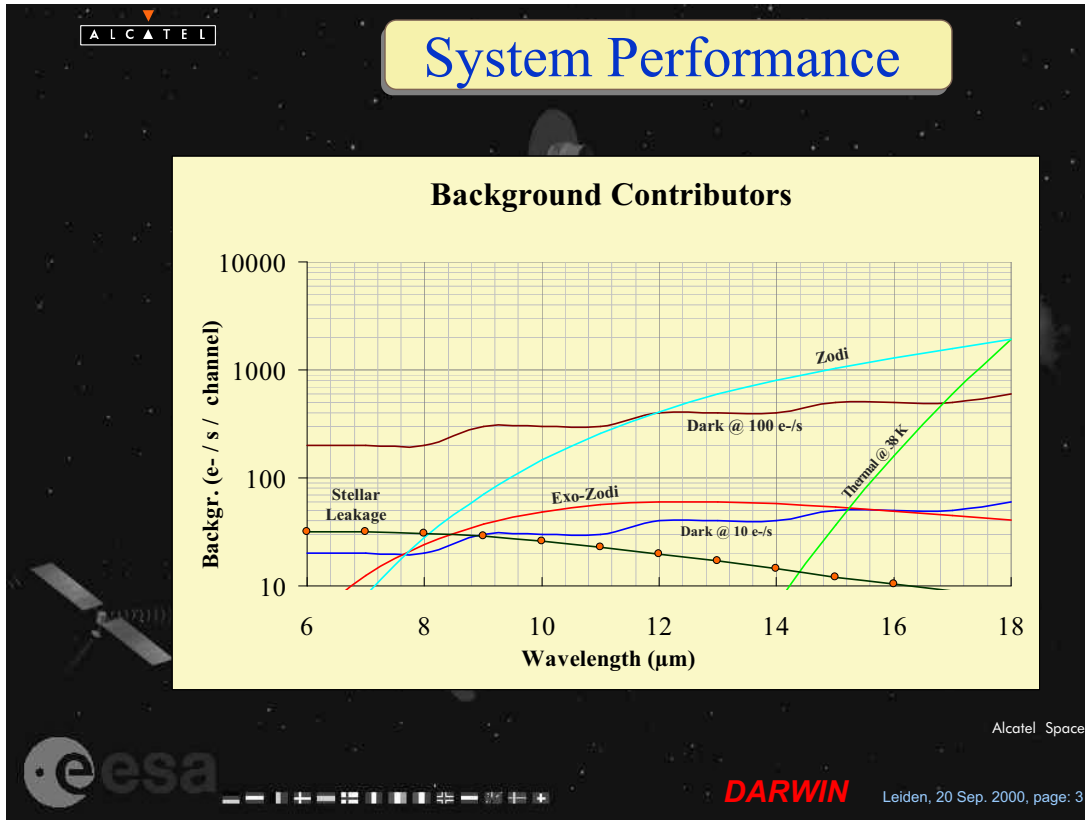
The need for nulling interferometry

- The problem →

The graph shows the spectral energy distribution of the Sun and Earth. The x-axis is $\log \lambda (\mu\text{m})$ ranging from -0.5 to 1.5. The y-axis is $\log \lambda, N_{\lambda}$ [photons m^{-2} s^{-1}] ranging from -2 to 10. The Sun's spectrum is shown as a solid black line, and the Earth's spectrum is shown as a dashed black line. A red arrow indicates a difference of $7 \cdot 10^6$ photons m^{-2} s^{-1} at $\log \lambda = 1.0$. The Earth's spectrum shows absorption features for O_2 , H_2O , O_3 , and CO_2 .

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Comparison of stellar & planetary signals

Earth + Sun at 10 pc, $l = 10$ mm, $l/Dl = 20$

	$e^- s^{-1} channel^{-1}$
Planet	0.14
Solar Zodi	140
Exo-Zodi	45
Stellar leaks	$30 * 10^5 / \rho$

⇒ background noise must be controlled

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But, how do you get the null ???



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Bracewell : 2 pupils

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Other nulling configurations

Concept :	Bracewell	Angel's Cross	5-telescope "Darwin"	Single DAC	Mariotti (monoplane)	OASES	Modified Mariotti (multi-plane / hub)
Characteristics ↓							
Rej. Law	θ^2	θ^4	θ^4	θ^4	θ^4	θ^6	θ^4
Achromatic Phase delay	π	π	$2\pi/5$	π	π	π	π
Symmetric background rejection ?	✓	x	✓	x	✓	x	✓
Planet signal modulation	Rotation of interferometer	Rotation of interferometer	Rotation of interferometer	Rotation of interferometer	Internal modulation	Rotation of interferometer	Internal modulation
Beam combination complexity (1.... 5)	1	2	3	2	5	2	3
AOCS complexity (1.... 5)	1	2	3	2	5	2	3
Long delay lines ?	x	x	x	✓	✓	✓	x
Shortactive delay lines ?	✓	x	x	x	✓	✓	x
Imaging mode difficulty ?	5	3	2	4	2	3	1

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The InfraRed Space Interferometer - DARWIN

- **3-GAC nulling interferometer**
- Hexagonal configuration of 6 telescopes
- Central hub / beam combiner
- Only integral π phase shifts needed ...
 - ⇒ unequal amplitude sharing (1/3, 2/3, 3/3)
- Internal modulation

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GAC : Fresnel vector diagram

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Beam Combination (1)

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Beam Combination (2)

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Beam Combination (3)

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Transmission map & nulling

Surface brightness (M_{λ}^2/m^2)

orbital radius (au)

Stellar disk

planet

exo - ZL

local ZL + other background signals

Surface brightness (M_{λ}^2/m^2)

orbital radius (au)

Transmission map profile

orbital radius (au)

θ^4 rejection law

On-axis null

Surface brightness (M_{λ}^2/m^2)

orbital radius (au)

Planet signal

Field Stop

Residual stellar leakage

Detected ZL & other background flux

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Connected Structure?

Free Flyer	Connected structure
Variable baseline (≤ 1 km)	Fixed baseline (≤ 100 m?)
2D / 3D configurations	Linear configurations
No mechanical coupling	Low frequency resonance
High cost	Very high cost
New technology	Technology available (ISS, SIM)

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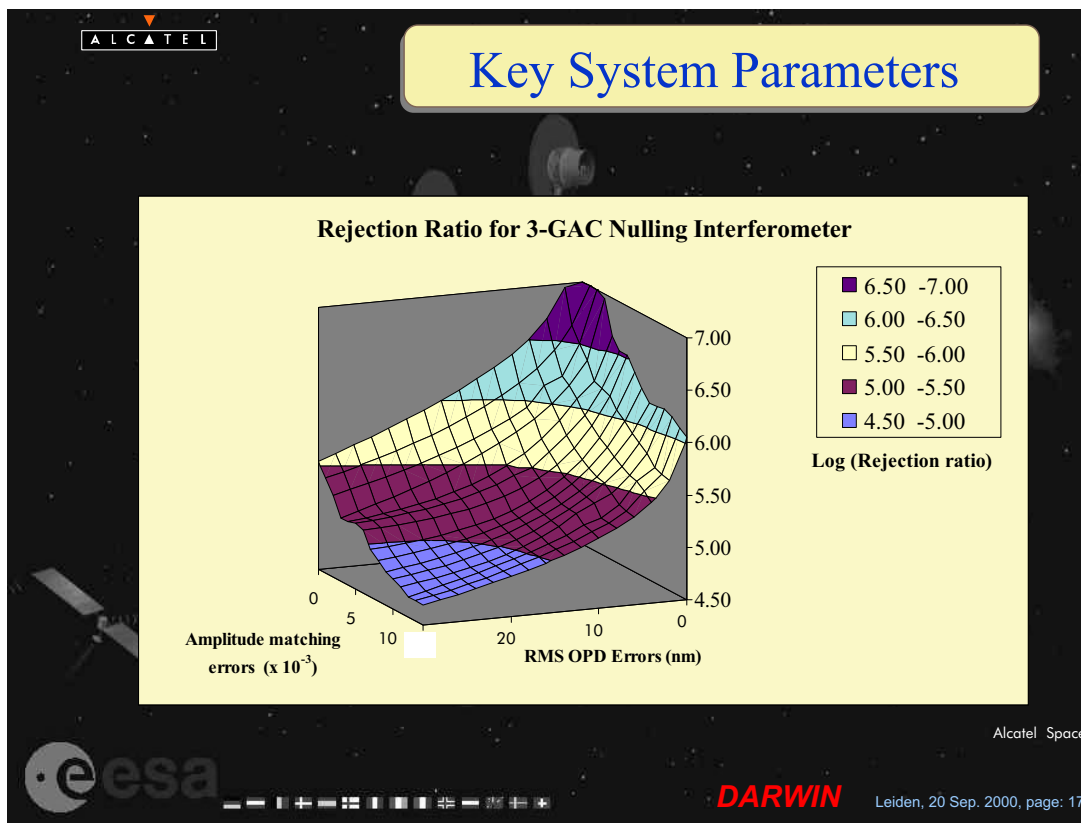
Key System Parameters

- Optical nulling of "on axis" star by $\sim 10^5$
- Baseline accuracy : 1 cm
- Optical path difference : 20 nm
- Telescope pointing : 24 mas
- Amplitude matching : 10^{-2}

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Technical challenges

- Stellar light rejection to $\sim 10^{-5}$
- Distinguish between exo-planets and exo-zodi :
 - rotation of the complete interferometer during observations ?
 - internal modulation ?
- Thermal IR observations :
 - telescopes and structures at ~ 40 °K
 - detectors cooled to ~ 6 °K
- System complexity :
 - complex control systems
 - complex beam combining optics
 - redundancy, cost, reliability

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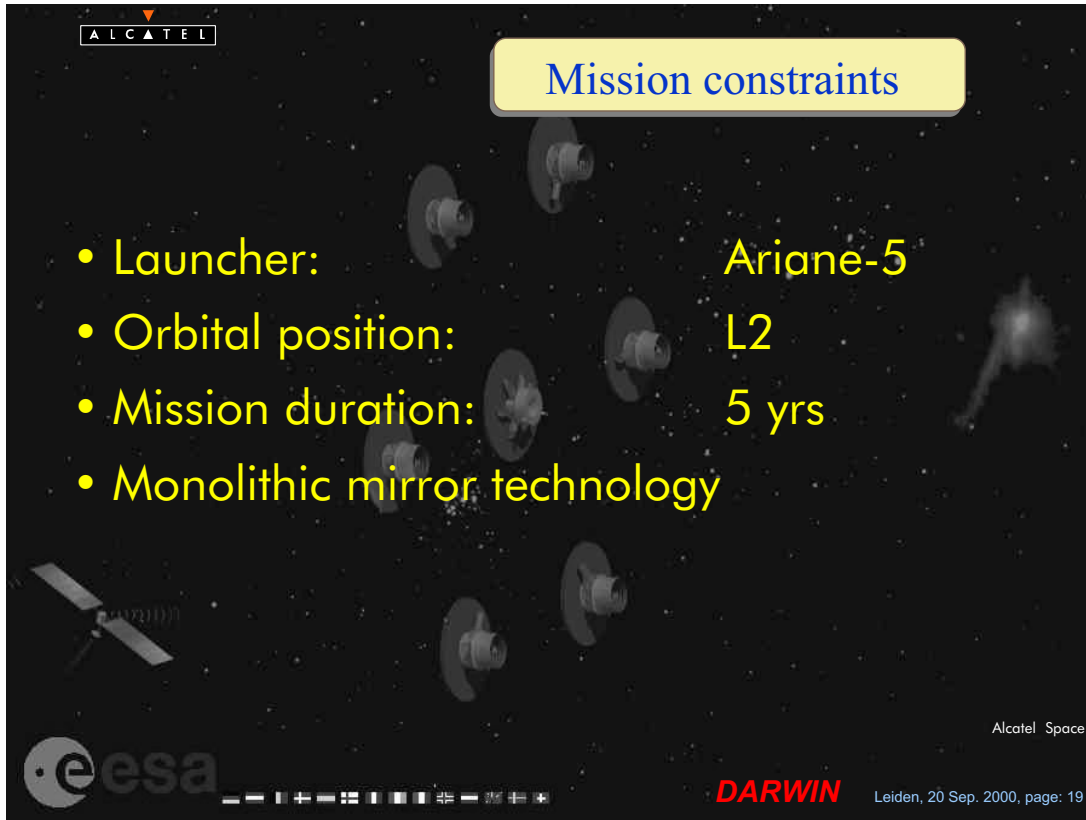
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Mission constraints

- Launcher: Ariane-5
- Orbital position: L2
- Mission duration: 5 yrs
- Monolithic mirror technology



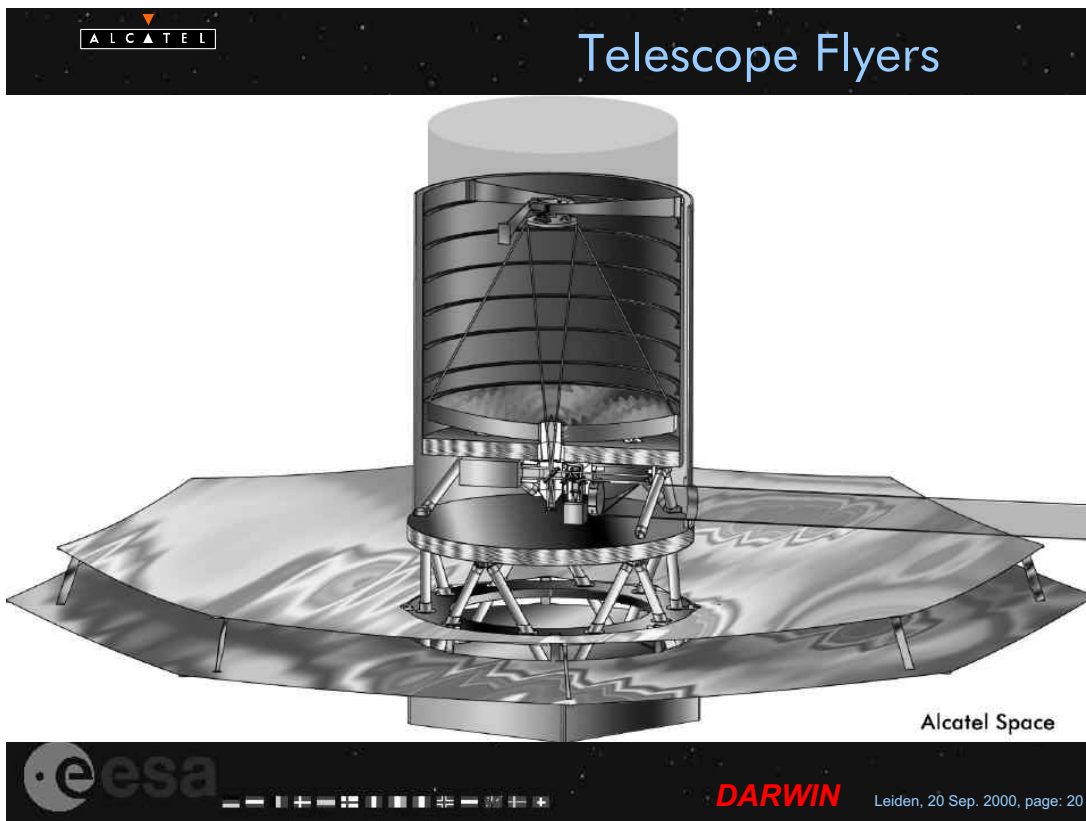
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Telescope Flyers



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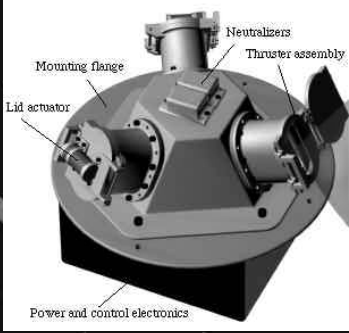
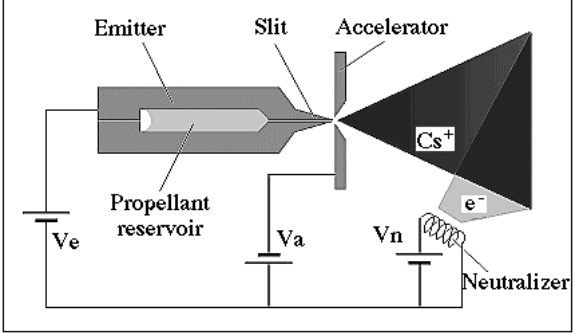
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Propulsion

- Fine control: μN - thrust
- Coarse control: mN - thrust
- FEED - Field Emission Electric Propulsion
- Cold gas ?

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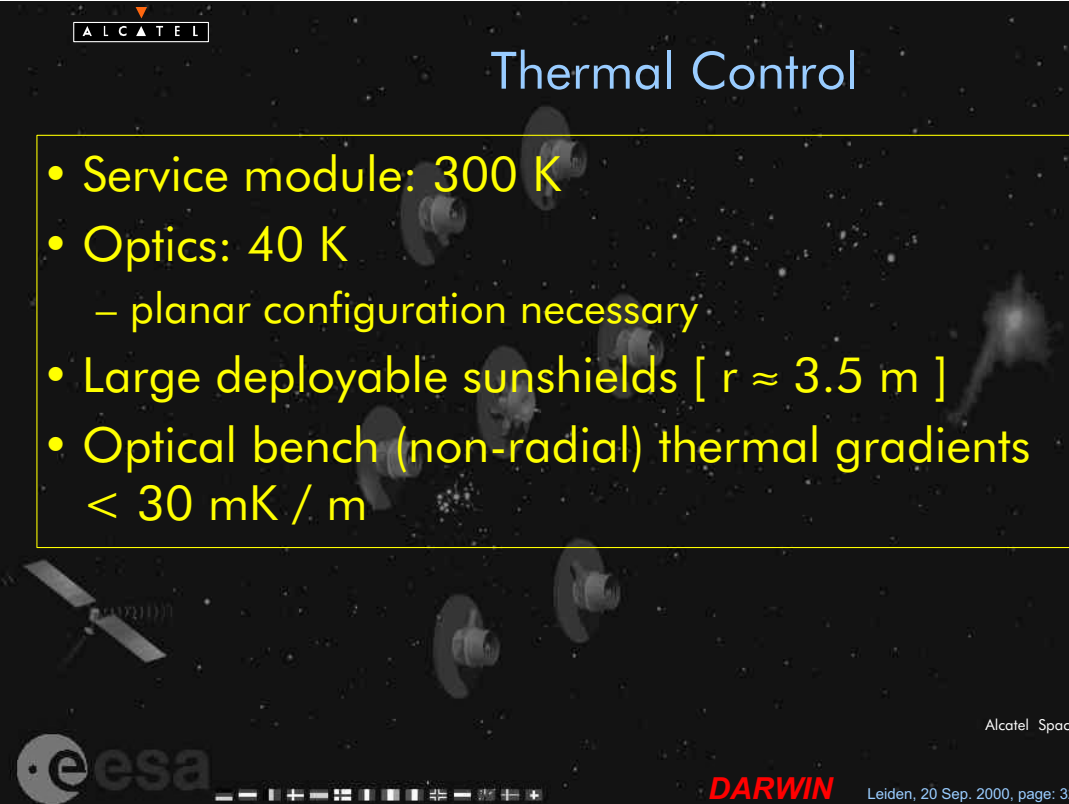
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Thermal Control

- Service module: 300 K
- Optics: 40 K
 - planar configuration necessary
- Large deployable sunshields [$r \approx 3.5 \text{ m}$]
- Optical bench (non-radial) thermal gradients $< 30 \text{ mK / m}$



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Communications

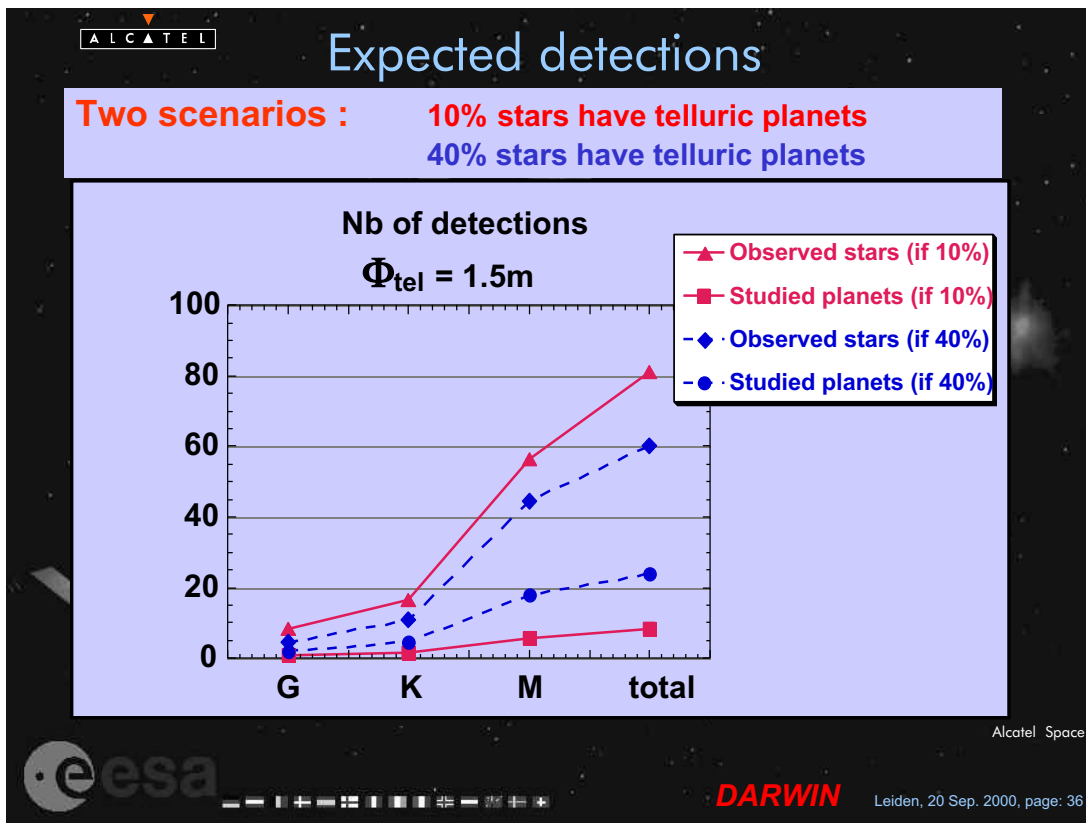
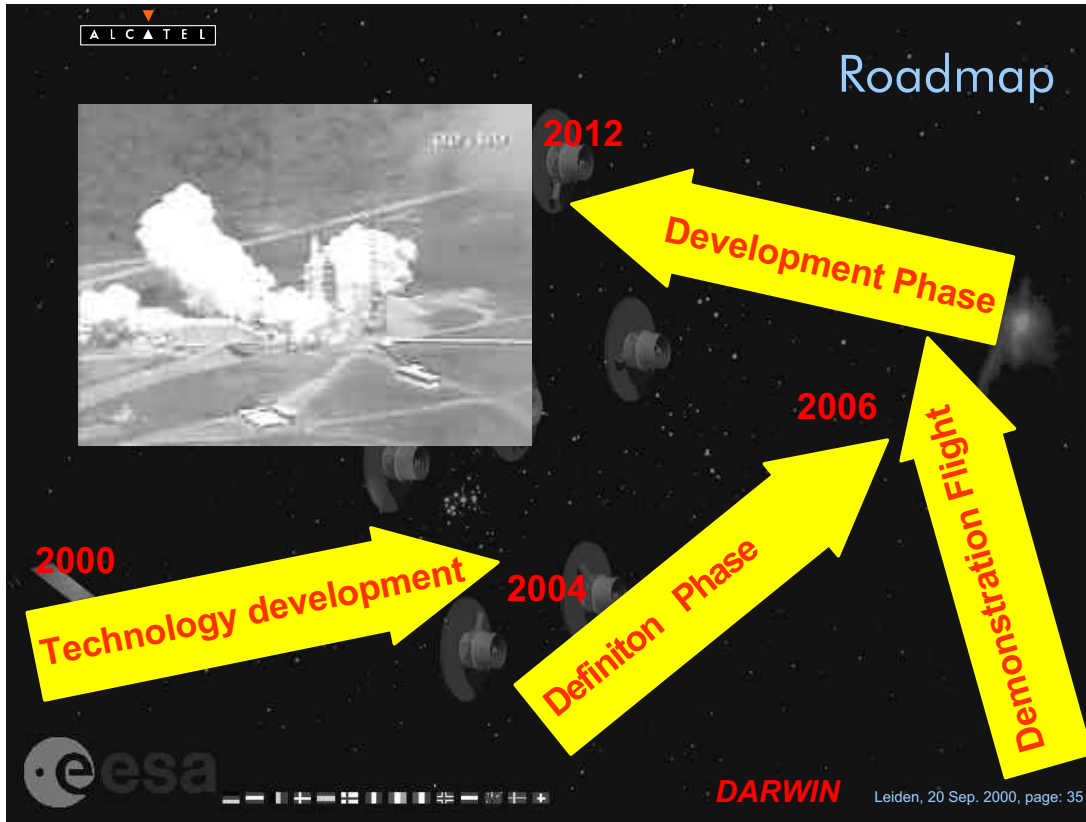
- X-band ground links
 - Command rate: 4 kbps
 - Telemetry rate: 20 kbps
- S-band inter-satellite
- Emergency S-band links

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Launcher Accomodation

- Beamcombiner
- Telescope flyers
- Upper dispenser
- Master Satellite
- Telescope flyers
- Lower dispenser

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Technology development requirements **Optics**

– **Major sub-systems**

- **Science telescope** : very fast parabolic primary mirror (F/1), with ultra-lightweight structure, WFE requirement of ~ 50 nm.
- **WFC** : more detailed analysis needed, to check no major difficulties.
- **Fringe tracker** : design and detailed analysis needed.
- **Flux matching device** : design has been proposed, real life performance yet to be evaluated.

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Technology development requirements **Optics**

– **Optical components**

- Precise manufacturing control of dichroic mirrors, beam-splitters & compensating plates : 10^{-3} uniformity in transmission - from one component to any other, at all wavelengths.
- High uniformity of refractive index of thermal IR materials (dispersion $\sim 10^{-3}$).
- Precise control of spectral response (dielectric coatings) in both polarisations, for dioptric components.
- Monomode optical fibres for thermal IR ($6 \sim 18 \mu\text{m}$), with suitable homogeneity and good optical throughput. Chromatic requirements TBD.
- Characterisation of nominal performance of the former at 40 K.
- Achromatic Pi phase shifter : differential chromatism ~ 1 nm ($\sim \leq 2\pi/1000$ in phase).

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Technology development requirements **OPD & Pointing Control**

- **FEEP Thrusters - science mode**
 - Noise requirement : $< 1.65 \mu\text{N} @ 1 \text{ Hz}$.
 - Thrust resolution : $< 3 \mu\text{N}$
 - Maximum thrust : $> 50 \mu\text{N}$
- **FEEP Thrusters - interferometer reconfiguration**
 - Maximum thrust : $\sim \text{mN}$
- **RF ranging & goniometry systems**
- **Comparative laser metrology system**
- **Transverse motion sensor**

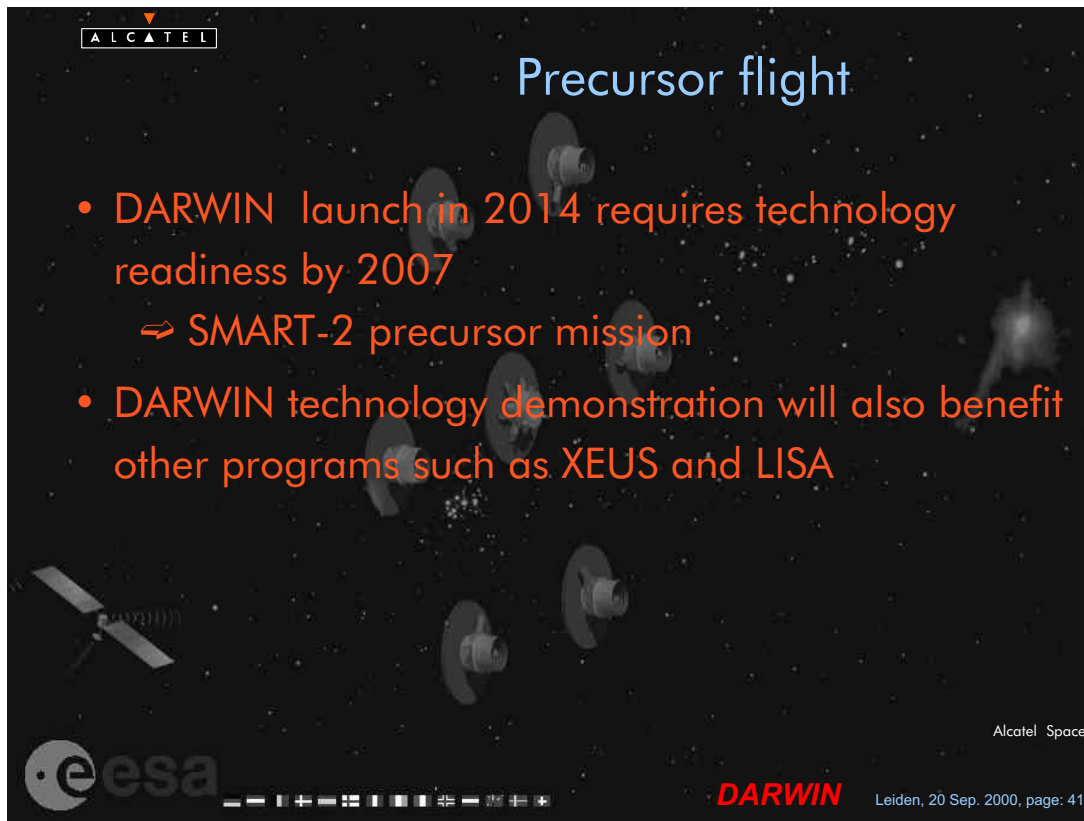
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The need to go to space versus ground based development

- Nulling demonstration in lab and at VLTI
- Fringe detection in space is a key element of the precursor mission
 - Fringe detection is an integral part of the control loop for formation flying

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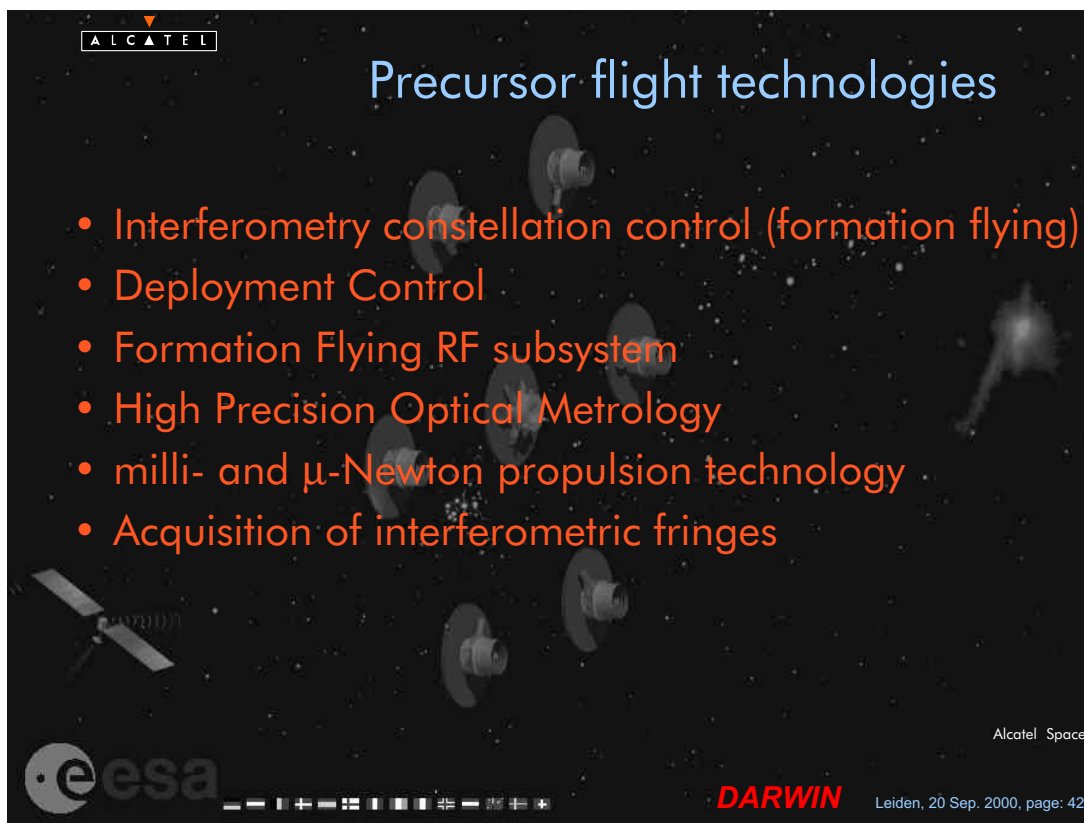


Precursor flight

- DARWIN launch in 2014 requires technology readiness by 2007
 - ⇒ SMART-2 precursor mission
- DARWIN technology demonstration will also benefit other programs such as XEUS and LISA

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Precursor flight technologies

- Interferometry constellation control (formation flying)
- Deployment Control
- Formation Flying RF subsystem
- High Precision Optical Metrology
- milli- and μ -Newton propulsion technology
- Acquisition of interferometric fringes

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International collaboration

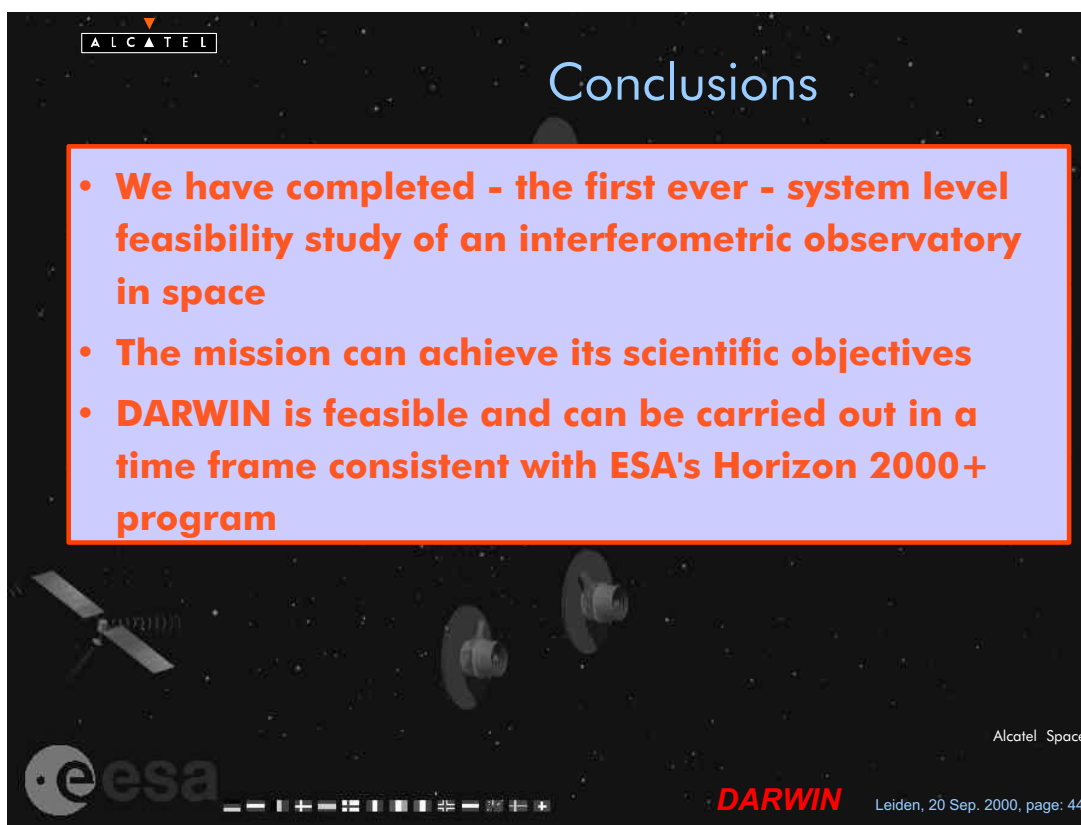
Close co-operation exists at the scientific level between ESA and NASA on DARWIN and TPF

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Conclusions

- We have completed - the first ever - system level feasibility study of an interferometric observatory in space
- The mission can achieve its scientific objectives
- DARWIN is feasible and can be carried out in a time frame consistent with ESA's Horizon 2000+ program

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